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► **To cite this version:**

Jin Wu, Zebo Zhou, Hassen Fourati, Yuhua Cheng. A Super Fast Attitude Determination Algorithm for Consumer-Level Accelerometer and Magnetometer. IEEE Transactions on Consumer Electronics, Institute of Electrical and Electronics Engineers, 2018, 64 (3), pp.375 - 381. <10.1109/tce.2018.2859625>. <hal-01922922>

HAL Id: hal-01922922

<https://hal.inria.fr/hal-01922922>

Submitted on 14 Nov 2018

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A Super Fast Attitude Determination Algorithm for Consumer-Level Accelerometer and Magnetometer

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Abstract—A super fast attitude solution is obtained for consumer electronics accelerometer-magnetometer combination. The quaternion parameterizing the orientation is analytically derived from a least-square optimization that maintains very simple form. Like previously developed approaches, this algorithm does not require predetermined magnetometer reference vector. It has been proven in the paper that the proposed algorithm is equivalent to two recent representative methods. Computational complexity analysis shows that the proposed algorithm has the least floating-number operations. Comparisons with recent and classical methods indicate the definite superiority of the proposed algorithm on execution time in real embedded applications.

Index Terms—Attitude determination, analytical solution, accelerometer, magnetometer, quaternion.

I. INTRODUCTION

THE MICRO-ELECTRO-MECHANICAL-SYSTEMS (MEMS) accelerometer-magnetometer combination is a key component in many consumer-level products requiring motion sensing [1]. These applications have enriched the functionalities of nowadays intelligent devices like cellphones, smart wearables, gait logger, unmanned aerial vehicles and etc. [2]–[4]. The accelerometer and magnetometer along with gyroscope can together form a general inertial measurement unit (IMU) [2]. In principle, using accelerometer and magnetometer respectively, one device can compute the attitude solution by means of local gravity and magnetic-field sensing [5]. For most consumer-level equipments, such sensor combination is sufficient for evaluation of orientation measurements [6], [7].

For nowadays accelerometer and magnetometer, there are always algorithms estimating Euler angles for motion detection on the chip. This characteristic is usually not

recommended for long-endurance products because of their disadvantage on high power consumption. Actually, a faster sensor fusion method leads to less power consumption and higher system reliability [8]. Existing methods calculating the attitude results have been developed in the last decade. The most representative ones are factored quaternion algorithm (FQA) and Markley’s two-vector method, which both employ the geometric relationship between sensing axes [9], [10]. Some other algorithms are proposed recently, e.g., algebraic quaternion algorithm (AQUA, [11]), symbolic accelerometer-magnetometer fusion [12], singular-value-decomposition (SVD) filter [13] and etc. [14].

The attitude determination results are often later processed by complementary and Kalman filters [15]–[17] for more smooth estimates. This requires a later loosely coupled fusion with gyroscope [18]. Generally speaking, to achieve the goal of extremely low power consumption, one attitude estimation algorithm should have at least 1) Energy-saving sensors; 2) Time-costly measurement equations; 3) Simple and explicit solutions. For instance, replacing gyroscopes with accelerometer and magnetometer, the power consumption can be significantly reduced accordingly since the MEMS gyroscopes are usually driven with more electricity. Recently, Makni *et al.* [19] have developed an orientation algorithm using the adaptive Kalman filter where the gyroscope measurements are configured to be intermittent so that the power can be saved. The left two factors are also very important to reduce computation burdens. For example, the Kalman filter also consumes much more computational resources than simple ones like complementary filters. Therefore for many embedded products, the complementary filters are usually employed for saving power.

In previous research, the fusion problem of accelerometer and magnetometer is typically solved via sophisticated mathematical results. The aforementioned representatives for accelerometer and magnetometer, e.g., FAQ, AQUA and SVD do not show very good performance for long-endurance application according to complicated internal computations, e.g., matrix manipulations, trigonometric functions, square roots and etc. Following the steps of these former endeavors, in this paper we have the following main contributions.

- 1) A novel fast quaternion solution is derived according to the least-square optimization.
- 2) The proposed attitude determination algorithm is systematically investigated showing that it owns the same accuracy as previous representatives achieve.

Manuscript received March 14, 2018; revised May 26, 2018 and July 17, 2018; accepted July 19, 2018. Date of publication July 25, 2018; date of current version September 24, 2018. This work was supported in part by the National Natural Science Foundation of China under Grant 41604025, and in part by the State Key Laboratory of Geodesy and Earths Dynamics (Institute of Geodesy and Geophysics, Chinese Academy of Sciences) under Grant SKLGED2018-3-2-E. (*Corresponding author: Zebo Zhou.*)

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Digital Object Identifier 10.1109/TCE.2018.2859625

- 3) The numerical analysis is presented which indicates the super fast computation speed of the proposed algorithm. The experiments also indicate that the new method has an evident improvement of over 50% on the execution time.

This paper is organized as follows: Section II states the problem and presents the solution. Section III discusses the equivalence of various algorithms. Section IV presents the computation complexities of representative algorithms and the proposed one. Section V contains experimental validations while Section VI gives concluding remarks.

II. PROBLEM STATEMENT AND SOLUTION

The accelerometer and magnetometer have their normalized observations $\mathbf{A}^b = (a_x, a_y, a_z)^T$, $\mathbf{M}^b = (m_x, m_y, m_z)^T$ respectively in the body frame b . Along with their normalized reference vectors $\mathbf{A}^r = (0, 0, 1)^T$, $\mathbf{M}^r = (m_N, 0, m_D)^T$ in reference frame r (here is the North-East-Down) such that

$$\begin{cases} a_x^2 + a_y^2 + a_z^2 = 1 \\ m_x^2 + m_y^2 + m_z^2 = 1 \\ m_N^2 + m_D^2 = 1 \end{cases} \quad (1)$$

they are related by the direction cosine matrix (DCM) $\mathbf{C} \in SO(3)$ with the following Wahba's optimization:

$$\arg \min_{\substack{\mathbf{C}\mathbf{C}^T = \mathbf{C}^T\mathbf{C} = \mathbf{I} \\ \det(\mathbf{C}) = +1}} \left\{ w \|\mathbf{A}^b - \mathbf{C}\mathbf{A}^r\|^2 + (1-w) \|\mathbf{M}^b - \mathbf{C}\mathbf{M}^r\|^2 \right\} \quad (2)$$

where w is the weight of accelerometer and $1-w$ belongs to magnetometer [12]. The Wahba's problem can be solved with many existing numerical methods, e.g., QUEST, FLAE, ESOQ and etc. [20]. It is also noted that the optimization is currently solved by means of numerical methods like gradient-descent algorithm (GDA, [21], [22]), Gauss-Newton algorithm (GNA, [23]), Levenberg-Marquardt algorithm (LMA, [24]). However, according to uncertainties of these algorithms on the convergence and execution time consumption, we do not address discussion and comparison of them in this paper. By a series of transformation, it is proven that Wahba's solution is equivalent to finding the eigenvector of the maximum eigenvalue associated with the Davenport's \mathbf{K} matrix [25]:

$$\mathbf{K} = \begin{bmatrix} \mathbf{B} + \mathbf{B}^T - \text{tr}(\mathbf{B})\mathbf{I} & \mathbf{z} \\ \mathbf{z}^T & \text{tr}(\mathbf{B}) \end{bmatrix} \quad (3)$$

where $\text{tr}(\dots)$ is the matrix trace; \mathbf{I} stands for the identity matrix. Other parametric values are provided as follows

$$\begin{aligned} \mathbf{B} &= w\mathbf{A}^b(\mathbf{A}^r)^T + (1-w)\mathbf{M}^b(\mathbf{M}^r)^T \\ \mathbf{z}^T &= (B_{23} - B_{32}, B_{31} - B_{13}, B_{12} - B_{21}) \end{aligned} \quad (4)$$

in which B_{ij} stands for the element of \mathbf{B} in i -th row and j -th column. For the specific fusion of accelerometer and magnetometer, the above mentioned methods will not obtain results within relative execution time. The reason is that these methods are general solvers. They compute the analytical forms of the characteristic polynomial of \mathbf{K} with many existing results [8]. The eigenvector can be computed until the

maximum eigenvalue having been found out via numerical iterations, e.g., Gauss-Newton recursion. Therefore, we are thinking how to derive the closed-form eigenvalue of this problem and solve the optimal eigenvector directly.

Inserting vector components into the above equations, the analytic form of \mathbf{K} can be computed. Then the characteristic polynomial is obtained by

$$\det(\mathbf{K} - \lambda_{\mathbf{K}}\mathbf{I}) = 0 \quad (5)$$

where

$$\lambda_{\mathbf{K}}^4 + \tau_1\lambda_{\mathbf{K}}^2 + \tau_2\lambda_{\mathbf{K}} + \tau_3 = 0 \quad (6)$$

in which the parameter set can be determined by [26]

$$\begin{aligned} \tau_1 &= -2\text{tr}^2(\mathbf{B}) + \text{tr}[\text{adj}(\mathbf{B} + \mathbf{B}^T)] - \mathbf{z}^T\mathbf{z} \\ \tau_2 &= -\text{tr}[\text{adj}(\mathbf{K})] \\ \tau_3 &= \det(\mathbf{K}) \end{aligned} \quad (7)$$

where adj denotes the adjoint matrix. In advance, the coefficients can be computed to

$$\begin{aligned} \tau_1 &= -4w^2 + 4w - 2 + 4\alpha m_D w(w-1) \\ \tau_2 &= 0 \\ \tau_3 &= 4\alpha m_D w(w-1) \left(-2w^2 + 2w - 1 \right) \\ &\quad + 4w^2(w-1)^2(\alpha^2 - m_N^2) \\ &\quad + 4w^4 - 8w^3 + 8w^2 - 4w + 1 \end{aligned} \quad (8)$$

with which the eigenvalues of \mathbf{K} is given by

$$\begin{aligned} \lambda_{\mathbf{K},1} &= \sqrt{(1-w)^2 + w^2 + 2w(1-w)(\alpha m_D + V)} \\ \lambda_{\mathbf{K},2} &= \sqrt{(1-w)^2 + w^2 + 2w(1-w)(\alpha m_D - V)} \\ \lambda_{\mathbf{K},3} &= -\sqrt{(1-w)^2 + w^2 + 2w(1-w)(\alpha m_D - V)} \\ \lambda_{\mathbf{K},4} &= -\sqrt{(1-w)^2 + w^2 + 2w(1-w)(\alpha m_D + V)} \end{aligned} \quad (9)$$

where

$$\begin{aligned} \alpha &= a_x m_x + a_y m_y + a_z m_z \\ V &= m_N \sqrt{1 - \alpha^2} \end{aligned} \quad (10)$$

The derivation details are given in the Appendix A. From the above equation, we have the identity $V > 0$ since the magnetic dip angle $\theta \in [-\frac{\pi}{2}, \frac{\pi}{2}]$ ensures $m_N = \cos\theta > 0$ [9]. Sorting these eigenvalues in descending order, it is obtained that

$$\lambda_{\mathbf{K},1} \geq \lambda_{\mathbf{K},2} > \lambda_{\mathbf{K},3} \geq \lambda_{\mathbf{K},4} \quad (11)$$

Therefore the attitude quaternion should be the eigenvector associated with the eigenvalue $\lambda_{\mathbf{K},1}$.

In accelerometer-magnetometer fusion, in fact the magnetic dip angle is not required before sensor setup, since [25], [27]

$$m_D = \alpha, m_N = \sqrt{1 - \alpha^2} \quad (12)$$

always holds, with the constraint of the DCM. In this way, the roll and pitch are not affected by magnetic sensing and the

analytic form of the maximum eigenvalue can be simplified to

$$\begin{aligned}\lambda_{\mathbf{K},1} &= \sqrt{(1-w)^2 + w^2 + 2w(1-w)(\alpha m_D + m_N \sqrt{1-\alpha^2})} \\ &= \sqrt{(1-w)^2 + w^2 + 2w(1-w)(m_D^2 + m_N^2)} \\ &= 1\end{aligned}\quad (13)$$

Inserting the above eigenvalue back into $\mathbf{K}-\mathbf{I}$, the final results of the row echelon form can be given by

$$\mathbf{K}-\mathbf{I} \Rightarrow \begin{pmatrix} 1 & \cdots & a \\ & 1 & \vdots & b \\ & & 1 & c \\ & & & d \end{pmatrix} \quad (14)$$

where

$$\begin{aligned}a &= \frac{1}{P} \left\{ m_y[m_D - a_x m_N + \alpha(w-1) - w m_D] \right. \\ &\quad \left. + a_y[m_N m_x + m_N^2(1-w) + w - w \alpha m_D] \right\} \\ b &= \frac{1}{P} \left\{ m_N m_z + \alpha m_x(1-w) + m_D m_x(w-1) \right. \\ &\quad \left. + a_z m_N[m_D(w-1) - \alpha w] \right. \\ &\quad \left. + a_x[m_N^2(w-1) - w + w \alpha m_D] \right\} \\ c &= \frac{1}{P} \{ m_N[-(1+a_z)m_y + a_y(m_D + m_z + w\alpha - w m_D)] \} \\ d &= \frac{1}{P} \{ 2(1 + \alpha^2 - 2\alpha m_D - m_N^2)(1-w)w \} \\ P &= -1 - m_N m_x - a_z m_N m_x + (w-1)a_z m_N^2 \\ &\quad + (w-1)m_D m_z - a_z w + a_x m_N(m_D + m_z - m_D w) \\ &\quad + \alpha(m_D + m_z + a_z m_D w + a_x m_N w - m_z w)\end{aligned}\quad (15)$$

As $\alpha = m_D$, the fundamental solution to $(\mathbf{K}-\mathbf{I})\mathbf{q} = \mathbf{0}$ is given by

$$\begin{aligned}\tilde{\mathbf{q}} &= (\tilde{q}_0, \tilde{q}_1, \tilde{q}_2, \tilde{q}_3)^T \\ \tilde{q}_0 &= (a_z - 1)(m_N + m_x) + a_x(m_D - m_z) \\ \tilde{q}_1 &= (a_z - 1)m_y + a_y(m_D - m_z) \\ \tilde{q}_2 &= a_z m_D - a_x m_N - m_z \\ \tilde{q}_3 &= -a_y(m_N + m_x) + a_x m_y\end{aligned}\quad (16)$$

which shows that the solution is free of pre-determined weights of vector observations. Note that the solved quaternion is a vector-wise one, i.e., [28]

$$\tilde{\mathbf{q}} = \left(\mathbf{n} \sin \frac{\phi}{2}, \cos \frac{\phi}{2} \right)^T = (\tilde{q}_0, \tilde{q}_1, \tilde{q}_2, \tilde{q}_3)^T \quad (17)$$

where \mathbf{n} is the rotation axis and ϕ is the rotation angle about the rotation axis. For common use, one should convert it to regular form. Hence the final unitary quaternion solution is given by

$$\mathbf{q} = \left(\cos \frac{\phi}{2}, \mathbf{n} \sin \frac{\phi}{2} \right)^T = \frac{(\tilde{q}_3, \tilde{q}_0, \tilde{q}_1, \tilde{q}_2)^T}{\sqrt{\tilde{q}_0^2 + \tilde{q}_1^2 + \tilde{q}_2^2 + \tilde{q}_3^2}} \quad (18)$$

III. EQUIVALENCE ANALYSIS

Equation (16) is in fact a extremely simplified solution to the Wahba's problem. Therefore, combining Markley's two-vector approach or other Wahba's solutions with (12) gives the same numerical solution of attitude quaternion shown in (16).

The overwhelming factor to obtain the proposed simple solution presented in (16) is indeed (12). In this section we are going to show that for representative existing magnetometer reference vector free algorithms, this preliminary always holds. Let us review the computation procedures of FQA and AQUA [9], [11]. The FQA extracts the roll and pitch via the accelerometer measurements using elementary quaternions:

$$\text{pitch} : \begin{cases} a_x = \sin \theta \\ \mathbf{q}_{pitch} = (\cos \frac{\theta}{2}, 0, \sin \frac{\theta}{2}, 0)^T \end{cases} \quad (19a)$$

$$\text{roll} : \begin{cases} \sin \varphi = -\frac{a_y}{\cos \frac{\theta}{2}} \\ \cos \varphi = -\frac{a_z}{\cos \frac{\theta}{2}} \\ \mathbf{q}_{roll} = (\cos \frac{\varphi}{2}, \sin \frac{\varphi}{2}, 0, 0)^T \end{cases} \quad (19b)$$

where θ, φ denote the pitch and roll angles respectively. \mathbf{q}_{roll} and \mathbf{q}_{pitch} are independently estimated quaternions for two rotation directions roll and pitch. AQUA solves the joint quaternion of roll and pitch by an indefinite system of equations using acceleration attitude kinematics:

$$\begin{cases} 2(q_1 q_3 + q_0 q_2) = a_x \\ 2(q_2 q_3 - q_0 q_1) = a_y \\ q_0^2 - q_1^2 - q_2^2 + q_3^2 = a_z \end{cases} \quad (20)$$

This leads to the general solution of

$$\mathbf{q}_{acc} = \begin{cases} \left(\kappa_1, -\frac{a_y}{2\kappa_1}, \frac{a_x}{2\kappa_1}, 0 \right)^T, & a_z \geq 0 \\ \left(-\frac{a_y}{2\kappa_2}, \kappa_2, 0, \frac{a_x}{2\kappa_2} \right)^T, & a_z < 0 \end{cases} \quad (21)$$

where

$$\begin{aligned}\kappa_1 &= \sqrt{\frac{a_z + 1}{2}} \\ \kappa_2 &= \sqrt{\frac{1 - a_z}{2}}\end{aligned}\quad (22)$$

Therefore the first stages of estimating the non-yaw angles belonging to FQA and AQUA respectively are very similar and they have the same accuracy in principle. Then the two algorithms all rotate the magnetometer measurements into an intermediate earth frame avoiding using local magnetic reference vector. Differently, they are performed by means of quaternion multiplication

$$\begin{pmatrix} 0 \\ \tilde{\mathbf{M}} \end{pmatrix} = \mathbf{q}_{pitch} \otimes \mathbf{q}_{roll} \otimes \begin{pmatrix} 0 \\ \mathbf{M}^b \end{pmatrix} \otimes \mathbf{q}_{roll}^{-1} \otimes \mathbf{q}_{pitch}^{-1} \quad (23)$$

for FQA. And DCM rotation by AQUA, such that

$$\mathbf{C}^T(\mathbf{q}_{acc})\mathbf{M}^b = \tilde{\mathbf{M}} \quad (24)$$

where \otimes denotes the quaternion product; $\tilde{\mathbf{M}} = (h_x, h_y, h_z)^T$ is the temporary magnetometer reference vector lying on the intermediate frame. The yaw quaternion is then estimated by satisfying

$$\left(0, \sqrt{h_x^2 + h_y^2}, 0, h_z \right)^T = \mathbf{q}_{yaw} \otimes \begin{pmatrix} 0 \\ \tilde{\mathbf{M}} \end{pmatrix} \otimes \mathbf{q}_{yaw}^{-1} \quad (25)$$

or

$$\mathbf{C}^T(\mathbf{q}_{mag})\tilde{\mathbf{M}} = \left(\sqrt{h_x^2 + h_y^2}, 0, h_z \right)^T \quad (26)$$

TABLE I
FLOPS OF ALGORITHMS

Algorithm	+	×	/	$\sqrt{\dots}$	$\cos(\dots)$
Proposed SAAM	18	16	1	2	0
FQA	18	53	4	3	6
Markley	33	51	3	4	0
AQUA	24	38	2	4	0

The final full-attitude quaternion is formed by quaternion multiplication and singularity avoidance. This shows that FQA and AQUA are the same in principle. Note that in fact

$$\left(\sqrt{h_x^2 + h_y^2}, 0, h_z\right)^T = \mathbf{M}^r \quad (27)$$

A key step of the above algorithms is to compute $\tilde{\mathbf{M}}$. From (26), one can find out that

$$\mathbf{C}^T(\mathbf{q}_{mag})\mathbf{C}^T(\mathbf{q}_{acc})\mathbf{M}^b = \left(\sqrt{h_x^2 + h_y^2}, 0, h_z\right)^T \quad (28)$$

Note that

$$\mathbf{C}^T(\mathbf{q}_{mag})\mathbf{C}^T(\mathbf{q}_{acc}) = \mathbf{C}^T(\mathbf{q}) \quad (29)$$

is actually the full-attitude rotation allowing for $\mathbf{M}^b = \mathbf{C}(\mathbf{q})\mathbf{M}^r$. This shows that the FQA and AQUA are essentially the same with Wahba's optimization for accelerometer and magnetometer. Then it is ensured that (12) holds for these methods as well.

Yun's FQA and Valenti's AQUA are both algebraic methods free of magnetometer reference vectors. In general, AQUA and FQA extracts the maximum information from accelerometer and magnetometer measurements, which has been systematically verified in last sub-section. From the presented approach shown above, we can see that the proposed method is consistent with AQUA and FQA on this point owning the internal dynamics of (12). Hence, the proposed algorithm, Markley's two-vector method, AQUA and FQA have the same attitude determination accuracy. As they employ quite different architectures, the computation speeds vary a lot. The computation complexity analysis and execution time evaluation will be presented in the following sections.

IV. COMPUTATION COMPLEXITY

The proposed method owns the simplest form of quaternion as we know. Then the algorithm is named as the Super-fast Attitude of Accelerometer and Magnetometer (SAAM). In this section, we are going to give statistics of floating-number operations (FLOPS) and exact attitude tests. Representatives like FQA, Markley's two-vector method and AQUA are introduced for comparisons. The FLOPS of various algorithms are presented in Table I.

For embedded platforms without floating-point number unit (FPU, [29]), floating number (IEEE-754 standard, [30]) operations are processed by the CPU. In such occasion, addition and multiplication are done instantaneously in one machine processing cycle. One floating-number square-root operation requires approximately 7 cycles while that of sine or cosine function are 15 cycles [31]. The division is usually 33%

TABLE II
TIME COMPLEXITIES OF ALGORITHMS

Algorithm	Without FPU	With FPU
Proposed SAAM	$O(49.5n)$	$O(35.5n)$
FQA	$O(188.0n)$	$O(83.0n)$
Markley	$O(116.5n)$	$O(85.5n)$
AQUA	$O(93.0n)$	$O(63.5n)$

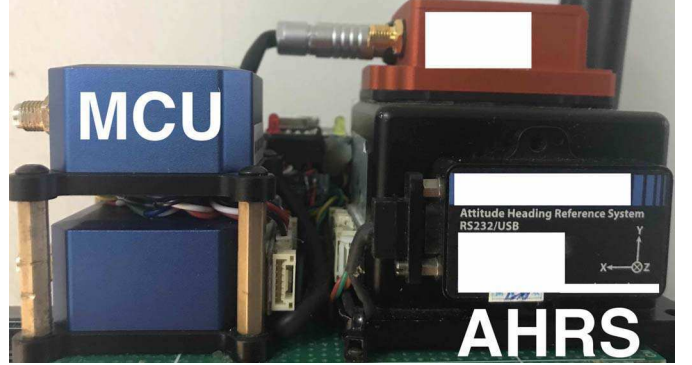


Fig. 1. Designed hardware for validation.

slower than addition or multiplication. When FPU takes place, the performance of square root and sine, cosine could be optimized using DSP for saving at most 95% computation time [32]. Now, let us define we have n loops to conduct. Then the time complexity of the algorithms with or without FPU can be calculated in Table II. Therefore, via the comparisons shown above, the theoretical algorithm advance in the speed is about 47.3% to 73.7% without FPU and 47.2% to 58.5% with FPU.

V. HARDWARE VALIDATION

The algorithms are executed on a commercial attitude and heading reference system (AHRS). The AHRS is connected to our 32-bit micro controller unit (MCU) with the processing unit of 400MHz core clock speed with FPU, via the serial port under the baudrate of 921600bps and sampling frequency of 500Hz. The internal digital low pass filter (DLPF) is tuned to be with cut-off frequency of 66.6Hz for default setting. All the numerical modules, e.g., square root and trigonometric functions are implemented using the public library offered by the hardware vendor. The hardware setup is shown in Fig. 1.

A record of raw sensor data and reference angles are logged with the MCU and the calculation is conducted on the chip with an embedded operating system called the FreeRTOS at the kernel scheduling frequency of 1000Hz. The scheduling method of the RTOS is set to the Round-Robin one that makes every task run synchronously [33]. Each algorithm is written into a task of scheduling frequency of 200Hz with the same priority. The computed attitude angles' root mean-squared errors (RMSEs) along with the statistics are shown in Fig. 2 and Table III. It is noted that the RMSE is computed with the reference Euler angles output from the AHRS.

Also, with the aid of the internal high resolution timer, the execution time consumption on chip is collected. The results

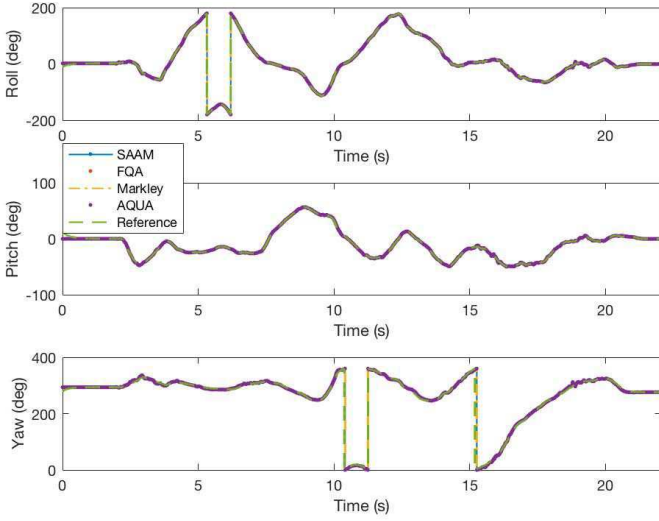


Fig. 2. The full-attitude test result.

TABLE III
RMSES OF ATTITUDE ANGLES

Algorithm	Roll (deg)	Pitch (deg)	Yaw (deg)
Proposed SAAM	7.008441954211032	1.190529228853902	24.732781316935142
FQA	7.008441954211161	1.190529228853916	24.732781316935156
Markley	7.008441954211033	1.190529228853914	24.732781316935142
AQUA	7.008441954211031	1.190529228853916	24.732781316935155

TABLE IV
EXECUTION TIME CONSUMPTION RESULT

Algorithm	Time (s)	Standard Deviation (s)
Proposed SAAM	1.77239×10^{-6}	4.70877×10^{-7}
FQA	1.18861×10^{-5}	1.40646×10^{-6}
Markley	9.10740×10^{-6}	1.01356×10^{-6}
AQUA	6.58190×10^{-6}	8.17700×10^{-7}

are summarized in Table IV where the standard deviation indicates the time consistency of a certain algorithm. From the results we can see that the proposed SAAM maintains the attitude determination accuracy and significantly reduces the computation time. The general speed of the SAAM is about 400 ~ 660% times that of existing algorithms. The test results on embedded computer basically coincide with the theoretical time complexity analysis in Section IV which verifies the correctness as well.

VI. DISCUSSION

The proposed SAAM is the fastest one compared with state-of-the-art algorithms. The saved CPU time can be used for less power, more effective scheduling, smarter failsafe of system and etc. As a matter of fact, from (16), we can see that it is the simple form that consumes the least execution time. This simple computation procedure also helps hardware engineers easily implement the circuit of this algorithm. Here, for example, when designing by means of digital circuit tools, some computations, e.g., square root and cosine functions are not easy to be implemented [34]. In the final stage of the algorithmic validation, we have designed the real-world integrated

circuit (IC). The current state of the circuit implementation indicates that much less electronic components are required with the proposed SAAM algorithm. The designed scheme has allowed for longer lifetime of devices in use without losing accuracy, according to its extreme low power consumption. Therefore, with the current obtained results, we hope that we may design a simple and cheap digital solution for combined attitude determination from gyroscope, accelerometer and magnetometer for more general requirements and mass production of IC in the future.

VII. CONCLUSION

In this paper, a novel solution to the attitude determination from accelerometer and magnetometer is proposed. We name it the SAAM algorithm, which employs the simplest quaternion form ever seen. Mandatory derivations are shown to derive the analytic solution. It is performed that the proposed SAAM is equivalent to FQA, AQUA and Markley's algorithm. The superiority of the algorithm is analysed via numerical evaluation of FLOPS and time complexity. The final experiment shows that the proposed SAAM has almost the same accuracy but consumes much less execution time, compared with representative algorithms. We hope that SAAM will boost the embedded attitude system design of accelerometer and magnetometer in the future.

APPENDIX A ANALYTICAL DERIVATIONS

Coefficients of Characteristic Polynomial: Using (7) and (1), we can compute the coefficients by

$$\begin{aligned} \tau_1 = & 4(w-1)wa_xm_Dm_x + 4(w-1)wa_y m_Dm_y \\ & + 4(w-1)wa_z m_Dm_z - 4w^2 + 4w - 2 \end{aligned} \quad (30a)$$

$$\tau_2 = 0 \quad (30b)$$

$$\begin{aligned} \tau_3 = & 1 - 8(w-1)w^3 a_x m_Dm_x - 8(w-1)w^3 a_y m_Dm_y \\ & - 8(w-1)w^3 a_z m_Dm_z + 8(w-1)w^2 a_x m_Dm_x \\ & + 8(w-1)w^2 a_y m_Dm_y + 8(w-1)w^2 a_z m_Dm_z \\ & - 4(w-1)wa_x m_Dm_x - 4(w-1)wa_y m_Dm_y \\ & - 4(w-1)wa_z m_Dm_z + 8(w-1)^2 w^2 a_x a_y m_x m_y \\ & + 8(w-1)^2 w^2 a_x a_z m_x m_z + 4(w-1)^2 w^2 a_z^2 m_y^2 \\ & + 4(w-1)^2 w^2 a_y^2 m_z^2 + 8(w-1)^2 w^2 a_y a_z m_y m_z \\ & + 8(w-1)^2 w^2 a_z^2 m_y^2 + 8(w-1)^2 w^2 a_z^2 m_z^2 \\ & - 4(w-1)^2 w^2 a_y^2 - 4(w-1)^2 w^2 a_z^2 \\ & - 4(w-1)^2 w^2 m_N^2 - 4(w-1)^2 w^2 m_y^2 \\ & - (w-1)^2 w^2 m_z^2 + 8w^4 - 16w^3 + 12w^2 - 4w \end{aligned} \quad (30c)$$

Repeating invoking (1) and (10) generates the results in (8).

APPENDIX B EIGENVALUES

The eigenvalues of \mathbf{K} can be solved via the roots of its characteristic polynomial. Such eigenvalues are give by

$$\lambda_{\mathbf{K},1,2,3,4} = \pm \sqrt{\xi_1 \pm \xi_2} \quad (31)$$

where

$$\xi_1 = 2 \sqrt{m_N^2 (w^2 - w)^2 \begin{pmatrix} a_x^2 m_y^2 + a_x^2 m_z^2 - 2a_x a_y m_x m_y \\ -2a_x a_z m_x m_z + a_y^2 m_x^2 + a_y^2 m_z^2 \\ -2a_y a_z m_y m_z + a_z^2 m_x^2 + a_z^2 m_y^2 \end{pmatrix}}$$

$$\xi_2 = a_x^2 w^2 - 2a_x m_D m_x w^2 + 2a_x m_D m_x w + a_y^2 w^2 - 2a_y m_D m_y w^2 + 2a_y m_D m_y w + a_z^2 w^2 - 2a_z m_D m_z w^2 + 2a_z m_D m_z w + m_D^2 m_x^2 w^2 - 2m_D^2 m_x^2 w + m_D^2 m_x^2 + m_D^2 m_y^2 w^2 - 2m_D^2 m_y^2 w + m_D^2 m_y^2 + m_D^2 m_z^2 w^2 - 2m_D^2 m_z^2 w + m_D^2 m_z^2 + m_N^2 m_x^2 w^2 - 2m_N^2 m_x^2 w + m_N^2 m_x^2 + m_N^2 m_y^2 w^2 - 2m_N^2 m_y^2 w + m_N^2 m_y^2 + m_N^2 m_z^2 w^2 - 2m_N^2 m_z^2 w + m_N^2 m_z^2 \quad (32)$$

Inserting (1) into it, we have

$$\xi_1 = 2 \sqrt{m_N^2 (w - 1)^2 w^2 \begin{bmatrix} a_x^2 (m_y^2 + m_z^2) \\ -2a_y m_y (a_x m_x + a_z m_z) \\ -2a_x a_z m_x m_z \\ + a_y^2 (m_x^2 + m_z^2) \\ + a_z^2 (m_x^2 + m_y^2) \end{bmatrix}}$$

$$\xi_2 = 1 - 2w - 2a_x m_D m_x w (w - 1) - 2a_y m_D m_y w (w - 1) - 2a_z m_D m_z w (w - 1) + 2w^2 \quad (33)$$

Combining

$$\begin{aligned} \alpha^2 &= (a_x m_x + a_y m_y + a_z m_z)^2 \\ &= a_x^2 m_x^2 + a_y^2 m_y^2 + a_z^2 m_z^2 + 2a_x a_y m_x m_y \\ &\quad + 2a_x a_z m_x m_z + 2a_y a_z m_y m_z \\ &= a_x^2 (1 - m_y^2 - m_z^2) + a_y^2 (1 - m_x^2 - m_z^2) \\ &\quad + a_z^2 (1 - m_x^2 - m_y^2) + 2a_x a_y m_x m_y \\ &\quad + 2a_x a_z m_x m_z + 2a_y a_z m_y m_z \\ &= 1 - a_x^2 (m_y^2 + m_z^2) - a_y^2 (m_x^2 + m_z^2) \\ &\quad - a_z^2 (m_x^2 + m_y^2) + 2a_x a_y m_x m_y \\ &\quad + 2a_x a_z m_x m_z + 2a_y a_z m_y m_z \end{aligned} \quad (34)$$

with the above equation (33) gives (9).

ACKNOWLEDGMENT

The source code of the paper has been uploaded to <https://github.com/zarathustr/SAAM> for public evaluation.

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